TYPE-CERTIFICATE DATA SHEET

No. EASA.A.586

F260

Type Certificate Holder:

LEONARDO S.p.A. – Divisione Velivoli

Piazza Monte Grappa, 4

00195 Rome
Italy

Airworthiness Category: Small Aeroplanes

For Model: F260
F260B
F260C
F260D
F260E
F260F
SF260TP

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SECTION A: F260

A.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 19 June 1963

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 16 March 1966

A.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 19 June 1963

2. **Airworthiness Requirements:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** N.A.

A.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel.

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**

Span:	8,350 m
Length:	7,100 m
Height:	2,410 m

5. **Engine:**

5.1.1 Model	Lycoming O-540-E4A5
5.1.2 Type Certificate	State of Design Engine TCDS No. FAA n° E-295 EASA Engine TCDS No: N.A
5.1.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4 g
Acrobatic	-3 / +6 g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-1B/8467-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), Minimum allowable for repairs 1,905 m (75 in.) Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total Usable fuel 235 lt (62 US Gal)
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.)
8.3 Coolant	N.A.

10. **Airspeed limits** (See Note 3)**:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	162 Kts (187 mph) IAS
Maximum Flap Extended Speed	VFE:	108 Kts (125 mph) IAS
Maximum Gear Extended Speed	VLE:	108 Kts (125 mph) IAS

For detailed information see approved Aircraft Flight Manual.

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights** (See Note 3)**:**

Utility	Maximum take-off / landing weight	1102 Kg (2430 lbs)
Acrobatic	Maximum take-off / landing weight	1000 Kg (2205 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50m forward of firewall.

16. **Control Surface Deflections** (See Note 3)**:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+20° / -25° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity** (See Note 3)**:**

Utility	2 (two passengers) (1 at +2,550m 1 at +3,300m)
Acrobatic	1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:**

Utility	41Kg at +3,800m
Acrobatic	No baggage allowed

21. **Wheels and Tyres** See Flight and Maintenance Manual for Tyres and Wheels information

22. **Serial Numbers Eligible** From S/N 1-01 to 2-75.

Aviamilano NC 502 and 503 (See Note 3)

A.IV. Operating and Service Instructions

1. **Flight Manual, Document No** (See Note 3)**:** F.260 and F.260 B RAI-Approved Flight Manual 65.665/T dated April 18, 1968, Revision 1 through 7 and subsequent approved revisions.

2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.
F260-2 Maintenance Manual (Not RAI Approved).

3. **Service Letters and Service Bulletins:** As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.

4. **Required Equipment:** Refer to equipment list in Flight Manual.

A.V. Notes

1. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.

   The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 4 lbs at +105,5 in.
	Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

2. Aircraft Model F.260 with Serial Numbers 2-54 and up incorporate wings modified outline P/N 260-01-76, as model F.260B.

3. For Aviamilano NC 502 and 503, approved on March 16, 1966, the above mentioned limitations shall be replaced with the following ones:

Air Speed:		
– Never Exceed Speed VNE:	220 Kts (252 mph)	
Maximum weight:	850 Kg Acrobatic Category	
Number of seats:	1 at +100 in. – Acrobatic Category	
Minimum Equipment:	See Flight Manual	
Flight Manual:	RAI-Approved Flight Manual 49.785/T dated March 15, 1966.	
Control Surface Deflections:	Flaps (U/D)	+0° / -45° ± 1 °
	Ailerons (U/D)	+24° / -12° ± 1 °
	Rudder (R/L)	+25° / -25° ± 1 °

As of 1969, the certificate holder changed its name from “Aviamilano – Costruzioni Aeronautiche, via M. Melloni 70, Milano” to “SIAI-Marchetti S.p.A. – Sesto Calende (VA)”.

SECTION B: F260B

B.I. General

1. **Data Sheet No:** EASA.A.586

2. **Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260B
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 05 February 1971

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 24 March 1971

B.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 05 February 1971

2. **Airworthiness Requirements:** Civil Air regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8.

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** N.A.

B.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260 except for vertical empennage of increased area and airfoil sections of wing leading edge).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**

Span:	8,350 m
Length:	7,100 m
Height:	2,410 m

5. **Engine:**

5.1.1 Model	Lycoming O-540-E4A5
5.1.2 Type Certificate	State of Design Engine TCDS No. FAA n° E-295 EASA Engine TCDS No: N.A
5.1.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4 g
Acrobatic	-3 / +6 g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-1B/8467-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), Minimum allowable for repairs 1,905 m (75 in.) Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total Usable fuel 235 lt (62 US Gal)
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.)
8.3 Coolant	N.A.

10. **Airspeed limits:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	162 Kts (187 mph) IAS
Maximum Flap Extended Speed	VFE:	108 Kts (125 mph) IAS
Maximum Gear Extended Speed	VLE:	108 Kts (125 mph) IAS

For detailed information see approved Aircraft Flight Manual.

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights**:

Utility	Maximum take-off / landing weight	1102 Kg (2430 lbs)
Acrobatic	Maximum take-off / landing weight	1000 Kg (2205 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50 m forward of firewall.

16. **Control Surface Deflections:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+20° / -25° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity**

Utility	2 (two passengers) (1 at +2,550m 1 at +3,300m)
Acrobatic	1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:**

Utility	41Kg at +3,800m
Acrobatic	No baggage allowed

21. **Wheels and Tyres** See Flight and Maintenance Manual for Tyres and Wheels information

22. **Serial Numbers Eligible** From S/N 3-76 and Subsequent.

B.IV. Operating and Service Instructions

1.	**Flight Manual, Document No:**	F.260 and F.260 B RAI-Approved Flight Manual 65.665/T dated April 18, 1968, Revision 1 through 7 and subsequent approved revisions.
2.	**Maintenance Manual, Document No:**	Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989. F260-2 Maintenance Manual (Not RAI Approved).
3.	**Service Letters and Service Bulletins:**	As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.
4.	**Required Equipment:**	Refer to equipment list in Flight Manual.

B.V. Notes

1. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
   The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 4 lbs at +105,5 in. Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

.

SECTION C: F260C

C.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260C
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 09 February 1976

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 23 October 1976

C.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 05 February 1971

2. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735;
EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** FAR 36 Appendix F

C.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260B except for lowered seats, ailerons with servotab, battery relocated forward and radio equipment).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**
Span: 8,350 m
Length: 7,100 m
Height: 2,410 m

5. **Engine:**

5.1 Basic
5.1.1 Model: Lycoming O-540-E4A5
5.1.2 Type Certificate: State of Design Engine TCDS No. FAA n° E-295
EASA Engine TCDS No: N.A
5.1.3 Limitations: For all operations, 2700 rpm (260 hp).

5.2 Additional

5.2.1 Model	Lycoming AEIO-540-D4A5
5.2.2 Type Certificate	State of Design Engine TCDS No. FAA n° 1E4 EASA Engine TCDS No: N.A
5.2.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4	g
Acrobatic	-3 / +6	g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-1B / 8467-8R Hartzell HC-C2YK-4F / FC8477-8R Hartzell HC-C2YK-1BF / F8477-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 for all propellers EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), No reduction permitted. Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total Usable fuel 235 lt (62 US Gal) with O-540-E4A5 Total Usable fuel 220,6 lt (58,3 US Gal) with AEIO-540-D4A5
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.) with O-540-E4A5 Usable 5,7 lt (6 qt.) with AEIO-540-D4A5
8.3 Coolant	N.A.

10. **Airspeed limits:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	162 Kts (187 mph) IAS
Maximum Flap Extended Speed	VFE:	108 Kts (125 mph) IAS
Maximum Gear Extended Speed	VLE:	108 Kts (125 mph) IAS

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights:**

Utility	Maximum take-off / landing weight	1102 Kg (2430 lbs)
Acrobatic	Maximum take-off / landing weight	1000 Kg (2205 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50m forward of firewall.

16. **Control Surface Deflections:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Ailerons servotab (U/D)	+23° / -14° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+20° / -25° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity**	Utility Acrobatic	2 (two passengers) (1 at +2,550m 1 at +3,300m) 1 (one passenger) (1 at +2,550m) (See Note a pag. 24).
20. **Baggage/cargo Compartments:**	Utility Acrobatic	41Kg at +3,800m No baggage allowed
21. **Wheels and Tyres**	See Flight and Maintenance Manual for Tyres and Wheels information	
22. **Serial Numbers Eligible**	From S/N 266 and subsequent.	

C.IV. Operating and Service Instructions

1. **Flight Manual, Document No:**	F.260C RAI-Approved Airplane Flight Manual 135.212/T dated October 20, 1976 and subsequent approved revisions. F.260C with Lycoming AEIO-540-D4A5 engine RAI-Approved Airplane Flight Manual 181.907/T dated May 11, 1983 and subsequent approved revisions. F.260C EASA - AFM Approval n° 10033166 dated 21-12-2010.
2. **Maintenance Manual, Document No:**	Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989. SF260C-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:**	As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.
4. **Required Equipment:**	Refer to equipment list in Flight Manual.

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C.V. Notes

1. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
   The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 4 lbs at +105,5 in. Wing tanks 26,9 lbs at +105,5 in. with AEIO-540-D4A5 engine only Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

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SECTION D: F260D

D.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260D
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 31 July 1980

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 14 December 1985

D.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 31 July 1980

2. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendments 23-7 for the paragraph 23-729, 23-735;
EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** FAR 36 Appendix F

D.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260C except for reinforced wing main spar).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**

Span:	8,350 m
Length:	7,100 m
Height:	2,410 m

5. **Engine:**

5.1 Basic

5.1.1 Model	Lycoming O-540-E4A5
5.1.2 Type Certificate	State of Design Engine TCDS No. FAA n° E-295 EASA Engine TCDS No: N.A
5.1.3 Limitations:	For all operations, 2700 rpm (260 hp).

5.2 Additional

5.2.1 Model	Lycoming AEIO-540-D4A5
5.2.2 Type Certificate	State of Design Engine TCDS No. FAA n° 1E4 EASA Engine TCDS No: N.A
5.2.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4	g
Acrobatic	-3 / +6	g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-1B/8467-8R Hartzell HC-C2YK-4F / FC8477-8R Hartzell HC-C2YK-1BF / F8477-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 for all propellers EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), No reduction permitted. Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total Usable fuel 235 lt (62 US Gal) with O-540-E4A5 Total Usable fuel 220,6 lt (58,3 US Gal) with AEIO-540-D4A5
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.) with O-540-E4A5 Usable 5,7 lt (6 qt.) with AEIO-540-D4A5
8.3 Coolant	N.A.

10. **Airspeed limits:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	174 Kts (200 mph) IAS
Maximum Flap Extended Speed	VFE:	108 Kts (125 mph) IAS
Maximum Gear Extended Speed	VLE:	108 Kts (125 mph) IAS

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights:**

Utility	Maximum take-off / landing weight	1100 Kg (2425 lbs)
Acrobatic	Maximum take-off / landing weight	1100 Kg (2425 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50m forward of firewall.

16. **Control Surface Deflections:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Ailerons servotab (U/D)	+23° / -14° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+20° / -25° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity**

Utility	2 (two passengers) (1 at +2,550m 1 at +3,300m)
Acrobatic	1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:**

Utility	41Kg at +3,800m
Acrobatic	No baggage allowed

21. **Wheels and Tyres** See Flight and Maintenance Manual for Tyres and Wheels information

22. **Serial Numbers Eligible** From S/N 563, 566, 568, 735, 739 to S/N 1999.

D.IV. Operating and Service Instructions

1. **Flight Manual, Document No:** F.260D RAI-Approved Airplane Flight Manual 215.709/T dated December 14, 1985 and subsequent approved revisions.
F.260D with Lycoming AEIO-540-D4A5 engine RAI-Approved Airplane Flight Manual 215.709/T dated December 14, 1985 and subsequent approved revisions.
F.260D EASA - AFM Approval n° 10033166 dated 21-12-2010

2. **Maintenance Manual, Document No:** Periodic Inspection List F.260 – F.260B – F.260C – F.260D Rev. 1 RAI Approved No 251.669/T dated February 7, 1989.
F260D/SF260D-2 Maintenance Manual (Not RAI Approved).

3. **Service Letters and Service Bulletins:** As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.

4. **Required Equipment:** Refer to equipment list in Flight Manual.

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D.V. Notes

1 Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 4 lbs at +105,5 in.
	Wing tanks 26,9 lbs at +105,5 in. with AEIO-540-D4A5 engine only
	Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

.

SECTION E: F260E

E.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260E
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 17 May 1991

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 21 January 1992

E.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 17 May 1991

2. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-961, 23-993, 23-1353 and including Amendment 23-7 for the paragraph 23-735, 23-955; Amendment 23-13 for the paragraph 23-1589; Amendment 23-18 for the paragraph 23-943, 23-959; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729, 23-991; Amendment 23-29 for the paragraph 23-994, 23-995; Amendment 23-34 for the paragraph 23-333, 23-863, 23-1581, 23-1583, 23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-40 for the paragraph 23-951; Amendment 23-41 for the paragraph 23-1309; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443;
EASA CS Part 23 Amendment 1, dated February 12, 2009, for parag. 23.1353 (h).

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** ICAO Annex 16, chapter 10.

E.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260D except for: reinforced wing with aerodynamic modifications for stall speed reduction, fuel system).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**

Span:	8,350 m
Length:	7,100 m
Height:	2,410 m

5. **Engine:**

5.1 Basic

5.1.1 Model	Lycoming IO-540-D4A5
5.1.2 Type Certificate	State of Design Engine TCDS No. FAA n° E-295 EASA Engine TCDS No: N.A
5.1.3 Limitations:	For all operations, 2700 rpm (260 hp).

5.2 Additional

5.2.1 Model	Lycoming AEIO-540-D4A5
5.2.2 Type Certificate	State of Design Engine TCDS No. FAA n° 1E4 EASA Engine TCDS No: N.A
5.2.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4	g
Acrobatic	-3 / +6	g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-4F / FC8477-8R Hartzell HC-C2YK-1BF / F8477-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 for all propellers EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), No reduction permitted. Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total usable fuel: 228,1 lt (60,26 US Gal) in:
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.) with IO-540-D4A5 Usable 5,7 lt (6 qt.) with AEIO-540-D4A5
8.3 Coolant	N.A.

10. **Airspeed limits:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	182 Kts (209 mph) IAS
Maximum Flap Extended Speed	VFE:	110 Kts (127 mph) IAS
Maximum Gear Extended Speed	VLE:	125 Kts (144 mph) IAS
Maximum Gear Operation Speed	VLO:	125 Kts (144 mph) IAS

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights:**

Utility	Maximum take-off / landing weight	1200 Kg (2645 lbs)
Acrobatic	Maximum take-off / landing weight	1200 Kg (2645 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50m forward of firewall.

16. **Control Surface Deflections:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+5° ± 30' / -30° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity**

Utility	2 (two passengers) (1 at +2,550m 1 at +3,300m)
Acrobatic	1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:**

Utility	41Kg at +3,800m
Acrobatic	No baggage allowed

21. **Wheels and Tyres** See Flight and Maintenance Manual for Tyres and Wheels information

22. **Serial Numbers Eligible** S/N 784 and from S/N 2001 to S/N 2999..

E.IV. Operating and Service Instructions

1. **Flight Manual, Document No:** F.260E RAI-Approved Airplane Flight Manual 281.522/SCMA dated May 29, 1992 and subsequent approved revisions.
F.260E EASA - AFM Approval n° 10033166 dated 21-12-2010

2. **Maintenance Manual, Document No:** F.260E RAI-Approved Airplane Maintenance Manual – Chap. 4 and Chap. 5 97/2262/MAE.
F260D/SF260D-2 Maintenance Manual (Not RAI Approved).

3. **Service Letters and Service Bulletins:** As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.

4. **Required Equipment:** Refer to equipment list in Flight Manual.

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E.V. Notes

1 Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 15 lbs at +105,5 in.
	Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

SECTION F: F260F

F.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** F260F
(c) **Variant**: --

3. **Airworthiness Category:** Utility
Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 17 May 1991

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 21 Jenuary 1992

F.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 17 May 1991

2. **Airworthiness Requirements:** Civil Air Regulation Part 3, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-951, 23-995, 23-1353 and including Amendment 23-7 for the paragraph 23-735; Amendment 23-13 for the paragraph 23-1589; Amendment 23-23 for the paragraph 23-345; Amendment 23-26 for the paragraph 23-729; Amendment 23-34 for the paragraph 23-333, 23-1581, 23-1583,23-1585, 23-1587; Amendment 23-36 for the paragraph 23-2; Amendment 23-38 for the paragraph 23-572; Amendment 23-42 for the paragraph 23-221, 23-341, 23-425, 23-443;
EASA CS Part 23 Amendment 1, dated February 12, 2009, for para. 23.1353 (h).

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** FAR Part 36 effective December 1, 1969 including Amendments 36-1 through 36-20, ICAO Annex 16, chapter 10...

F.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260D except for: reinforced wing with aerodynamic modifications for stall speed reduction).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.
4. **Dimensions:**

Span:	8,350 m
Length:	7,193 m
Height:	2,680 m

5. **Engine:**

5.1 Basic

5.1.1 Model	Lycoming O-540-E4A5
5.1.2 Type Certificate	State of Design Engine TCDS No. FAA n° E-295 EASA Engine TCDS No: N.A
5.1.3 Limitations:	For all operations, 2700 rpm (260 hp).

6. **Load factors:**

Utility	-2.2 / +4.4	g
Acrobatic	-3 / +6	g

7. **Propeller:**

7.1 Basic

7.1.1 Model	Hartzell HC-C2YK-4F / FC8477-8R Hartzell HC-C2YK-1BF / F8477-8R
7.1.2 Type Certificate	State of Design Propeller TCDS No. FAA P-920 for all propellers EASA Propeller TCDS No: N.A
7.1.3 Number of blades	2
7.1.4 Diameter	Max. 1,930 m (76 in.), No reduction permitted. Pitch at R = 0,762 m (30 in.), Max: + 32°, Min.: + 15° 30'
7.1.5 Sense of Rotation	Clockwise

8. **Fluids:**

7.1 Fuel	Aviation Grade duel minimum 91/96 or 100LL in accordance with latest issue of Service Instruction Lycoming N° 1070.
7.2 Oil	Oil with single or multi-viscosity aviation grade in accordance with latest issue of Service Instruction Lycoming N° 1014.
7.3 Coolant	N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel	Total fuel 243 lt (64 US Gal), stored in: 2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m 2 tip tanks of 72 lt (19 US Gal) each at 2,420 m Total Usable fuel 235 lt (62 US Gal)
8.2 Oil	11,4 lt (3 US Gal) at 0,800 m Usable 8,8 lt (9,25 qt.)
8.3 Coolant	N.A.

10. **Airspeed limits:**

Never Exceed Speed	VNE:	236 Kts (272 mph) IAS
Maximum Structural Cruising Speed	VNO:	187 Kts (215 mph) IAS
Design Manoeuvring Speed	VA:	182 Kts (209 mph) IAS
Maximum Flap Extended Speed	VFE:	110 Kts (127 mph) IAS
Maximum Gear Extended Speed	VLE:	125 Kts (144 mph) IAS
Maximum Gear Operating Speed	VLO:	125 Kts (144 mph) IAS

11. **Maximum Operating Altitude** N/A

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights:**

Utility	Maximum take-off / landing weight	1200 Kg (2645 lbs)
Acrobatic	Maximum take-off / landing weight	1200 Kg (2645 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50 m forward of firewall.

16. **Control Surface Deflections:**

Flaps (U/D)	+0° / -50° ± 1 °
Ailerons (U/D)	+24° / -13° ± 1 °
Elevator (U/D)	+24° / -16° ± 1 °
Elevator tab (U/D)	+5° ± 30' / -30° ± 1 °
Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity**

Utility	2 (two passengers) (1 at +2,550m 1 at +3,300m)
Acrobatic	1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:**	Utility Acrobatic	41Kg at +3,800m No baggage allowed
21. **Wheels and Tyres**	See Flight and Maintenance Manual for Tyres and Wheels information	
22. **Serial Numbers Eligible**	Applicable Serial Number: S/N 3001 to 3999.	

F.IV. Operating and Service Instructions

1. **Flight Manual, Document No:**	F.260F RAI-Approved Airplane Flight Manual 281.522/SCMA dated May 29, 1992 and subsequent approved revisions. F.260F EASA - AFM Approval n° 10033166 dated 21-12-2010
2. **Maintenance Manual, Document No:**	F.260F RAI-Approved Airplane Maintenance Manual - Chap 4. and Chap. 5 97/2262/MAE. F260D/SF260D-2 Maintenance Manual (Not RAI Approved).
3. **Service Letters and Service Bulletins:**	As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.
4. **Required Equipment:**	Refer to equipment list in Flight Manual.

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F.V. Notes

1 Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 4 lbs at +105,5 in. Tip tanks 8,6 lbs at +95,3 in.
Trapped engine oil:	5,07 lbs at +31,5 in.

SECTION G: SF260TP

G.I. General

1. **Data Sheet No:** EASA.A.586

2**. Type / Variant or Model**

(a) **Type:** F260
(b) **Model:** SF260TP
(c) **Variant**: --

3. **Airworthiness Category:** Acrobatic

4. **Type Certificate Holder:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome - Italy

5. **Manufacturer:** Leonardo S.p.A. - Divisione Velivoli Piazza Monte Grappa, 4 00195 Rome – Italy (See note m)

6. **National Application Date:** 30 July 1993

7. **National Certification Authority:** ENAC (C.O. A132 Rev. 15)

8. **National Certification Date:** 29 October 1993

G.II. Certification Basis

1. **Reference Date for determining the applicable requirements:** 30 July 1993

2. **Airworthiness Requirements:** Civil Air Regulation Part 23, dated May 15, 1956, including Amendments 3-1 through 3-8 and FAR Part 23 dated February 1, 1965, for the paragraph 23-251, 23-907, 23-1105 and including Amendment 23-7 for paragraph 23-729, 23-735, 23-937, 23-955, 23-1041, 23-1045, 23-1091, 23-1103, 23-1155, 23-1505, 23-1527; Amendment 23-14 for paragraph 23-173, 23-201, 23-572, 23-929, 23-1017, 23-1027; Amendment 23-15 for paragraph 23-951, 23-1013, 23-1015; Amendment 23-16 for paragraph 23-335; Amendment 23-17 for paragraph 23-787, 23-933, 23-977, 23-1111, 23-1165, 23-1303, 23-1309; Amendment 23-18 for paragraph 23-939, 23-943, 23-959, 23-1121, 23-1141, 23-1145, 23-1337; Amendment 23-20 for paragraph 23-1351(e); Amendment 23-21 for paragraph 23-45, 23-49, 23-65, 23-75(d), 23-77, 23-175, 23-1043, 23-1353, 23-1521; Amendment 23-23 for paragraph 23-863, 23-1545, 23-1557, 23-1583; Amendment 23-26 for paragraph 23-253, 23-361, 23-371, 23-991, 23-1305, 23-1529; Amendment 23-28 for paragraph 23-1549; Amendment 23-29 for paragraph 23-901, 23-903, 23-905, 23-995, 23-997, 23-1019, 23-1093, 23-1143, 23-1183, 23-1189; Amendment 23-31 for paragraph 23-629;

3. **Special Conditions:** N.A.

4. **Exemptions:** N.A.

5. **Deviations:** N.A.

6. **Equivalent Safety Findings:** N.A.

7. **Requirements elected to comply:** N.A.

8. **Environmental Standards:** FAR Part 36, App. F (Acoustic Certification)

G.III. Technical Characteristics and Operational Limitations

1. **Type Design Definition:** Refer to Report N° 598-260-059 Rev. / and subsequent approved revisions.

2. **Description:** Single engine low wing aircraft, propeller driven, side by side seat front seats and rear passenger, equipped with tricycle retractable landing gear. The airframe structure is an all-metallic construction utilising conventional aluminium and steel. (Same as Model F260D except for reinforced wing with aerodynamic modifications for stall speed reduction, and Allison Turboprop Model 250-B17D installation).

3. **Equipment:** Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the aircraft for the Airworthiness Certificate release.
Besides are required the following equipment:
Stall warning , Safe Flight Instruments Cop. 164S
Refer also to the Equipment list in FM

.

4. **Dimensions:**

Span:	8,350 m
Length:	7,400 m
Height:	2,410 m

5. **Engine:**

5.1.1 Model: Allison Gas Turbine 250-B17D

5.1.2 Type Certificate: State of Design Engine TCDS No. FAA E10CE
EASA Engine TCDS No: N.A

5.1.3 Limitations:

TO Power and Mac Cont.	320 SHP	
Propeller speed (Np):		
-TO and Max Cont Power	100%	2030 rpm.
-transient (15 sec.)	110%	2233 rpm.
Turbine Outside Temperature (TOT)		
-TO and Max Cont Power	810°C	
-transient (6 sec)	843°C	
-starting (10 sec)	810°C – 927°C	
Torque:		
-TO and Max Cont Power	78 PSI	78%

6. **Load factors:** Acrobatic -3 / +6 g

7. **Propeller:**

7.1 Basic

7.1.1 Model: Hartzell HC-B3TF-7A / T10173-25R

7.1.2 Type Certificate: State of Design Propeller TCDS No. FAA P15EA
EASA Propeller TCDS No: N.A

7.1.3 Number of blades: 3

7.1.4 Diameter: Max. 1,930 m (76 in.), No reduction permitted
Pitch at R = 0,762 m (30 in.), Flight Idle: +10° ± 0,5°

7.1.5 Sense of Rotation: Clockwise

8. **Fluids:**

7.1 Fuel: JP-4 o JP-5, according to MIL-T-5624; Jet A, Jet A1 o Jet B according to ASTM D-1655. Anti-icing additive, according to MIL-I-27686E, must be used in fuel not actually premixed for operation at ambient temperature below 5°C. The maximum concentration is authorized between 0,06% and 0,15%.

7.2 Oil: Oil in accordance with:
MIL-L-23699 above - 40°C (ambient temperature).
MIL-L-7808 below 0 °C (ambient temperature).

7.3 Coolant: N.A.

9. **Fluid capacities** (See Note 1)**:**

8.1 Fuel: Total fuel 243 lt (64 US Gal), stored in:
2 wing tanks of 49,5 lt (13 US Gal) each at 2,680 m
2 tip tanks of 72 lt (19 US Gal) each at 2,420 m
Total Usable fuel 220,6 lt (58,2 US Gal)

8.2 Oil: 8,7 lt (2,3 US Gal) at 0,800 m
Usable 6,8 lt (1,8 US Gal)

8.3 Coolant: N.A.

10. **Airspeed limits** (See Note 3)**:**

Maximum Operating Speed	VMO	229 Kts IAS (at 12000 ft) 196 Kts IAS (at 20000 ft) With linear variation for intermediate altitudes
Design Manoeuvring Speed	VP:	174 Kts IAS
Maximum Flap Extended Speed (20°)		125 Kts IAS
(50°)		108 Kts IAS
Maximum Gear Extended Speed	VLE:	108 Kts IAS
Maximum Gear Operation Speed	VLO:	108 Kts IAS

11. **Maximum Operating Altitude** 20000 ft

12. **Allweather Operations Capability** In standard configuration the aircraft is equipped and certificated for day VFR operations. Flights in known icing conditions is prohibited.

13. **Maximum Weights** (See Note 3)**:** Acrobatic Maximum take-off / landing weight 1160 Kg (2560 lbs)

14. **Centre of Gravity Range** See Flight Manual for CG envelope.

15. **Datum:** 1,50m forward of firewall.

16. **Control Surface Deflections:**	Flaps (U/D)	+0° / -50° ± 1 °
	Ailerons (U/D)	+24° / -13° ± 1 °
	Ailerons servo tab (U/D)	+23° / -14° ± 1 °
	Elevator (U/D)	+24° / -16° ± 1 °
	Elevator tab (U/D)	+20° / -25° ± 1 °
	Rudder (R/L)	+30° / -30° ± 1 °

17. **Levelling Means:** Two screws on the left side of fuselage.

18. **Minimum Flight Crew:** 1 pilot seated in the RH front seat. (1 at +2,550m)

19. **Maximum Passenger Seating Capacity** Acrobatic 1 (one passenger) (1 at +2,550m) (See Note a pag. 24).

20. **Baggage/cargo Compartments:** Acrobatic 20Kg at +3,300m (on rear seat location)

21. **Wheels and Tyres** See Flight and Maintenance Manual for Tyres and Wheels information

22. **Serial Numbers Eligible** S/N 661 and S/N 4001 and subsequent.

G.IV. Operating and Service Instructions

1. **Flight Manual, Document No:** SF.260TP RAI-Approved Airplane Flight Manual 93/3051/MAE dated October 28, 1993 and subsequent approved revisions.

2. **Maintenance Manual, Document No:** SF.260TP RAI-Approved Airplane Maintenance Manual Chap 4 and Chap 5 97/2262/MAE.
SF260TP-2 Airplane Maintenance Manual (Not RAI Approved).

3. **Service Letters and Service Bulletins:** As published by Leonardo S.p.A. - Divisione Velivoli and approved by Authority, as applicable.

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4. **Required Equipment:** Refer to equipment list in Flight Manual.

G.V. Notes

1. Current weight and balance report including list of equipment in the certificated empty weight, and loading instructions, must be provided for each aircraft at the time of original certification.
The certificated empty weight and corresponding center of gravity location must include:

Unusable fuel:	Wing tanks 9,26 lbs at +61,46 in.
	Tip tanks 29,32 lbs at +68,07 in.
Trapped engine oil:	8,82 lbs at +34,29 in.

NOTES :

a. For utility category, aircraft F260, F260B, F260C, F260D, rear seat may be occupied by two persons and/or baggage provided:
   - The total weight on rear seat (including baggage) is under 250 lb
   - The rear seat is equipped with two separate safety belts
   - Weight and G position are within limits
   - No baggage is on the baggage compartment
   - No radio equipment is in the radio compartment. (for F260C and F260 D)

b. The following placards must be displayed as indicated:
   - On Instrument Panel:
   - This airplane must be operated as a Utility or Acrobatic category airplane in compliance with Approved Airplane Flight Manual.
   - All markings and placards on this airplane apply to its operation as a Utility category airplane
   - For Acrobatic category operations, refer to Airplane Flight Manual.
   - All Placards required in the basic Approved Airplane Flight Manual, installed in the appropriate location.

c. Each individual airplane will be supplied with a placard that specifies the kind of operation such as VFR and IFR, Day or Night, to which the operation of the airplane is limited by the equipment installed.

d. The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. In addition, the following items of equipment are required: (all items notes as standard equipment in the Manufacturer's approved equipment list).
   - Pre-stall warning indicator, Safe Flight Indicator Corp. 164S.
   - Applicable F260() and SF260TP flight manuals. (See relative paragraphs "Operating and Service Instructions" of this document).

e. Up to November 29, 1989, SIAI Marchetti Spa

f. As of November 30, 1989, SIAI Marchetti Spa became a member of Agusta S.p.A. Group.

g. As of January 1, 1997, Aermacchi S.p.A. has acquired SIAI Marchetti S.r.L.

h. As of July 13, 2006, Aermacchi S.p.A. became Alenia Aemacchi S.p.A.

i. As of January 1, 2016, Alenia Aemacchi S.p.A became Finmeccanica S.p.A. – Divisione Velivoli.

l. As of January 1, 2017, Finmeccanica S.p.A. – Divisione Velivoli became Leonardo S.p.A. – Divisione Velivoli.

m. **Manufaturer chronology**

Up to 1968 "AVIAMILANO – Costruzioni Aeronautiche" (see note 3 of Chapter A – F260)
Up to November 29, 1989, SIAI Marchetti Spa
As of November 30, 1989, SIAI Marchetti Spa became a member of Agusta S.p.A. Group.
As of July 1st 1992 became SIAI Marchetti SrL
As of January 1, 1997, Aermacchi S.p.A. has acquired SIAI Marchetti S.r.L
As of July 13, 2006, Aermacchi S.p.A. became Alenia Aermacchi S.p.A.
As of January 1, 2016, Alenia Aemacchi S.p.A became Finmeccanica S.p.A. – Divisione Velivoli
As of January 1, 2017, Finmeccanica S.p.A. – Divisione Velivoli became Leonardo S.p.A. – Divisione Velivoli.

ADMINISTRATIVE SECTION

I. Acronyms & Abbreviations

AFM	Airplane Flight Manual
Amdt.	Amendment
AMM	Airplane Maintenance Manual
CG	Centre of Gravity
DWN	down
EASA	European Aviation Safety Agency
IAS	Indicated Airspeed
ICAO	International Civil Aviation Organization
kg	kilograms
km/h	kilometres per hour
MAC	Mean Aerodynamic Chord
N.A.	Not applicable
SC	Special Condition
TCDSN	Type Certificate Datasheet Noise
VFR	Visual Flight Rules

II. Type Certificate Holder Record

LEONARDO S.p.A. – Divisione Velivoli
Piazza Monte Grappa, 4
00195 Rome
Italy

III. Change Record

Issue	Date	Changes	TC issue
01	21/12/2011	Transfer to EASA Type Design	21/12/2011
02	09/09/2016	Typo page 11 corrected	09/09/2016
03	01/02/2017	Change of Type Certificate Holder Editorial changes.	24/01/2017